Amendments to the Specification:

Page 7, please amend the paragraph bridging pages 7 and 8 as follows:

Referring first to Figure 1, there is schematically shown therein a preferred deployable support structure 1 with dual reflectors for a spacecraft vehicle 2 embodying the present invention. The support structure 1 comprises a first arm carrier in the form of an articulated boom 5 having an antenna reflector 6 in deployed condition at its lower end 7 and a mount 8 for connecting the boom 5 to a section of a sidewall of the spacecraft 2 at its upper end. The arm carrier of the boom 5 has a shoulder joint 10, an elbow joint 11 and a wrist joint 12. Significantly, the ~~elbow joint 11~~ boom 5 has a dog-leg 17 to facilitate stowage of the reflector 6 against the spacecraft's sidewall. (The dog-leg is best seen in the second articulated boom 15 in Figure 1, and in the boom 5' in Figure 2.) The joints 10, 11, 12 are hingedly-connected at a plurality of points along the axial length of the arm carrier by means of a spring-operated mechanical hinge mechanism 13 or other arrangement, ~~(not shown)~~ such as an articulated stepper motor harmonic drive unit 13a, to permit pivotal movement of the arm carrier together with its associated reflector in directions perpendicular to its axis. It will be understood that the antenna reflector 6 is of a standard configuration (around 3.5m diameter) with long focal length capability.

Page 8, please amend the third paragraph, as follows:

As also shown, the stowed boom 15 with antenna reflector 16 is mounted against the spacecraft sidewall on a plurality of pyrotechnic hold-down points 23 (only two of which are visible in Figure 1). Each hold-down point is configured to allow compliance in certain directions to ensure that the boom and spacecraft structure do not impart unwanted thermal expansion loads on each other. It will be understood that the hold-down points are operably released prior to deployment of the boom/reflector.

Please amend the paragraph bridging pages 8 and 9, as follows:

In the described embodiment of Figure 1, the booms 5, 15 are about 7m long. By using booms of this length the feed structure 20 is mounted (as shown) directly onto the spacecraft top floor 21, alleviating the need for large heavy feed tower structures. The booms 5, 15 are formed of lightweight carbon fibre composite material. Because the boom structures are long, it will be understood that these structures extend some distance out into space from the spacecraft when deployed, and will become exposed to extreme temperatures, typically in the temperature range +140°C to - 180°C, during deployment. The inventors have recognised that this can cause problems with the boom joints between the hinge mechanisms and the carbon fibre composite (tube) sections. To address these problems, the boom joints are bolted using a metal bracket 22 with flexibility built into it so as to allow for change(s) in size between the materials.

Please amend the first full paragraph on page 9, as follows:

In operation of the thus described arrangement of Figure 1 it will be understood that the two reflectors 6, 16 can be deployed individually or sequentially from the same side of the spacecraft (using one boom per reflector). It will also be understood that antenna pointing can be provided to achieve this deployment by use of 2 axes APM (antenna pointing mechanism) or by use of articulated stepper motor harmonic drive units 13a. ~~(not shown).~~ In this embodiment, the APM 2 axes functions are mounted in the boom close to the reflector edge. This has the advantage of reducing/minimising the mass mounted on the reflectors whilst still providing a 2-axes geometry and allowing APM mass to be easily tied down for launch.